

ABSTRACT OF THE DISCLOSURE

An easily producible turbine nozzle for a gas turbine engine is provided, which is capable of preventing flutter of the turbine nozzle during operation of the gas turbine engine. The turbine nozzle comprises airfoils stacked along the stacking axis. The high curvature portions on suction surface in airfoil section successively formed along the stacking axis of the airfoil describe a parabola line that curves toward the pressure side of the airfoil when seen from the front or rear of the turbine nozzle. The high curvature portions on suction surface in airfoil section curve most at the center along the stacking axis of the airfoil from a straight line that connects with a first intersection between the parabola line and an inner band in the turbine nozzle and with a second intersection between the parabola line and an outer band in the turbine nozzle. The maximum curvature falls within a range from 0.02 to 0.03-fold of the stacking axis of the airfoil.